Amendments to the Specification

Please add the following new heading before paragraph [0001]:

BACKGROUND

Please replace paragraph [0001] with the following amended paragraph:

[0001] The present invention relates to a gas turbine vane, in particular a vane of an aircraft engine~~, according to the definition of the species in Patent Claim 1~~.

Please add the following new heading before paragraph [0005]:

SUMMARY OF THE INVENTION

Please replace paragraph [0005] with the following amended paragraph:

[0005] ~~On this basis, the~~ An object of the present invention is to create a novel gas turbine vane.

Please replace paragraph [0006] with the following amended paragraph:

[0006] ~~This object is achieved by a gas turbine vane according to Patent Claim 1.~~ According to the present invention, the suction side of the blade of the gas turbine vane has at least one micro-profiled or micro-structured area to optimize a flow around the suction side or the blade.

Please add the following new heading before paragraph [0010]:

BRIEF DESCRIPTION OF THE DRAWINGS

Please replace paragraph [0010] with the following amended paragraph:

[0010] Preferred refinements of the present invention arise from ~~the subclaims and~~ the following description. An exemplary embodiment of the present invention is, without being limited thereto ~~it~~, explained in greater detail based on the drawings.

Please add the following new heading before paragraph [0013]:

DETAILED DESCRIPTION

Please replace paragraph [0017] with the following amended paragraph:

[0017] Micro-profiled area 18 is assigned to a section of suction side 16 in which flow deceleration takes place. Starting at front edge 13 of blade 11 of gas turbine vane 10, first a flow acceleration and subsequently a flow deceleration take place on suction side ~~15~~ 16. The danger of flow separation exists in particular in the areas of suction side 16 of blade 11 in which the flow deceleration takes place. In terms of the present invention, micro-profiled or micro-structured area 18 is precisely situated in this section of suction side 16 in which a flow deceleration takes place. This area extends preferably over between 30% and 70%, in particular between 30% and 50%, of the profile depth of blade 11. It is thus assigned to a central area of suction side 16 of blade 11.

Please replace paragraph [0021] with the following amended paragraph:

[0021] In the area of suction side 16 of blade 11, gas turbine vane 10 of the exemplary embodiment shown in Figure 2 has a first area 20 on the side of the vane foot, which has a micro-profile or a micro-structure as set forth in the present invention. This first area 20 on the side of the vane footing is assigned to an area of blade 11 ~~which~~ that is exposed to high vibration stresses. Area 20 is preferably micro-profiled or micro-structured in such a way that blade 11 is strengthened in this area 20 and/or that compressive stresses are induced. This makes it possible to optimize the strength characteristics of blade 11 in addition to a positive effect on the flow around it.

Please replace paragraph [0024] with the following amended paragraph:

[0024] As ~~is apparent~~ shown in Figure 2, both micro-profiled or micro-structured areas 20 and 22 extend over a larger section of the profile depth than micro-profiled or micro-structured area 21.

Please replace paragraph [0025] with the following amended paragraph:

[0025] ~~According to the present invention, a~~ A gas turbine vane is thus proposed where at least one micro-profiled or micro-structured area is assigned to the suction side of the blade of the gas turbine vane. This micro-profiled or micro-structured area is assigned to that section of the suction side of the blade which is particularly at risk regarding a possible flow separation. The

present invention is based on the recognition that precisely this section of the suction side is only slightly stressed with regard to erosion or contamination, so that the micro-structured or micro-profiled area of the suction side of the blade retains its effectiveness even during operation of the gas turbine. In addition to optimizing the flow around the blade, the strength of the gas turbine vane may be positively affected with the aid of the micro-profiled or micro-structured areas. The improved flow around the gas turbine vanes, designed according to the present invention, results in a greater compression limit and thus in improved efficiency of the gas turbine.

Please amend the heading on top of page 6 as follows:

~~PATENT CLAIMS:~~ WHAT IS CLAIMED IS: